1. Approving Civil Aviation Authority Country	2.	3. Form tracking Number:
FAA/United States	**AUTHORIZED RELEASE CERTIFICATE** **FAA Form 8130-3, AIRWORTHINESS APPROVAL TAG**	**Repair Order #: 8169** **Revision #:**

4. Organization Name and Address:	5. Work order/Contract/Invoice
Rotor-Tech International 6700 CE Dixon Street, Stockton, CA 95206, Certificate No. R4TR418N P (209) 546-0380 Fax: (209) 546-0383	**Customer** **P.O. Number: R2021-77D**

6. Item:	7. Description:	8. Part Number:	9. Quantity:	10. Serial/Batch Number:	11. Status/Work:
1.	**Main Rotor Blade**	**C621A1006103**	**1**	**1982**	**Repaired**

12. Remarks:

ECF.120.MR.01.101	Inspection	EC 120 AMM 62-11-00, 6-1
ECF.120.MR.01.112	Replacement of Abrasion Strip - Position 1	ECF.120.MR.01-RD.01-5
ECF.120.MR.01.107	Caulking of Separations on SS Abrasion Strip 3, OB, Lower Surface	EC 120 AMM 62-11-00, 8-7
ECF.120.MR.01.102	Refinishing	EC 120 AMM 62-11-00, 8-1 ECF.SPM.RB.01.001
ECF.120.MR.01.105	Static Balance	ACS.SPM.RB.01.002

Rotor-Tech International certifies that the work specified in Block 11 & 12 was carried out in accordance with EASA Part 145, and in respect to that work, the component is considered ready to release to service under EASA Part 145 Approval No. EASA.145.6948.

TSN: 5430.9

13a. Certifies that the items identified above were manufactured in conformity to:

☐ Approved design data and are in condition for safe operation.

☐ Non-approved design data specified in block 12.

13b. Signature	13c. Approval Organization Number
13d. Name	13e. Date (dd/mmm/yyyy)

14a. ☒ 14 CFR 43.9 Return to Service ☒ Other regulation specified in Block 12

Certifies that unless otherwise specified in block 12, the work identified in block 11 and described in block 12 was accomplished in accordance with Title 14, Code of Federal Regulations, part 43 and in respect to that work, the items are approved for return to service

14b. Authorized Signature	14c. Approval/Certificate No:
	R4TR418N
14d. Name (Typed or Printed)	14e. Date (dd-mm-yyyy)
Ernesto Luna	**03/Feb/2022**

User/Installer Responsibilities

It is important to understand that the existence of this document alone does not automatically constitute authority to install the aircraft engine/propeller/article.

Where the user/installer performs work in accordance with the national regulations of an airworthiness authority different than the airworthiness authority of the country specified in Block 1, it is essential that the user/installer ensures that his/her airworthiness authority accepts aircraft engine(s)/propeller(s)/article(s) from the airworthiness authority of the country specified in Block 1.

Statements in Blocks 13a and 14a do not constitute installation certification. In all cases, aircraft maintenance records must contain an installation certification issued in accordance with the national regulations by the user/installer before the aircraft may be flown.

7. SUCCESSIVE STATUS, MAINTENANCE AND OVERHAUL OPERATIONS

Contractor	Date	Aircraft Version and S/N	Operation Unit (houres)			Reason for transfer (code and symptoms) - Accomplished work - Replaced parts Successive status before ACFT delivery - see front page of table 5
			Support	Partial	Total	
PAE Yuma	04/04/2018	EC120B S/N 1412	5430.2	236.6	4528.6	***MR T&B Prefromed Final Trim Tab Setting #6 +1.5 #7 +1.1. See JCN 179CB-2018-0053-0001***
PAE/Yuma	07/18/18	EC120B/SN/1412	5986.8	56.6	4585.2	T+B Completed Final TAB Setting TAB #6 +1.7 + #7 +1.2 See W.O. 179CB2018-I-0070-0003
PAE/Yuma	5/10/2019	EC120B/1412	5905.7	~~416.5~~	5004.1	MR T+B completed under WO: ~~37~~ 179CB 2019 I 0039 0001; final tab setting on tabs 6+7 are +4.1° + +4.2°. AP 3777058 IA
PAE Yuma	7/25/2019	EC120B/1412	5986.4	998.7	5084.8	Main rotor track and balance was completed under work order: 179CB-2019-I-0047-0002; final trim tab settings for main rotor blade s/n: 1982 were recorded as follows: tab #6: +4°, tab #7: +4° A&P 3394352 IA
Pae Yuma	4-3-20	EC120 1412	6332.5	346.1	5430.9	removed from 179CB due to erosion on leading edge causing perforation. w/o 179CB-2020-R-0016. AP 2759016 IA BLADE IS ASSIGNED TO AIRCRAFT 374HS FOR DISPOSAL
RTI	03/Feb/2022	----------			5430.9	/ Inspected and repaired IAW EC 120 AMM 62-11-00, 6-1, 8-1, 8-7, and RTI/ ACS approved procedures, removed and replaced abrasion strip position 1, repaired separation on SS abrasion strip 3, outboard edge, lower surface, refinished, and static balanced, RTI RO 8169, FAA Repair Station No. R4TR418N, EASA.145.6948, RTI 9

5-17-7
AC

1. Approving Civil Aviation Authority/Country: FAA/United States	2. AUTHORIZED RELEASE CERTIFICATE FAA Form 8130-3, AIRWORTHINESS APPROVAL TAG	3. Form Tracking Number: AE1721411

4. Organization Name and Address:	5. Work Order/Contract/ Invoice Number:
Airbus Helicopters, Inc. 2701 N. Forum Drive, Grand Prairie Texas 75052-7099 CRS VY1R575K	11000345810

6. Item:	7. Description:	8. Part Number:	9. Quantity:	10. Serial Number:	11. Status/Work:
1	MAIN ROTOR BLADE	C621A1006103	1	1982	Repaired

12. Remarks:

Maintenance Release: Airbus Helicopters, Inc. certifies that the part or component identified herein has been Repaired in accordance with Airbus Helicopters Aircraft Maintenance Manual: AMM 62-11-00, 8-8 Revision: 00200, including inspection and testing, and has determined that this part or component herein is airworthy as stated in block 14 dated: 09/May/2017.

Certifies that the work specified in block 11/12 was carried out in accordance with EASA Part-145 and in respect to that work the component is considered ready for release to service under EASA Part-145 Approval Number: EASA.145.5119.

TSN: 4086.1 TSO: Not Required Cycles: Not Required

13a. Certifies the items identified above were manufactured in conformity to:	14a. [X] 14 CFR 43.9 Return to Service [X] Other regulation specified in Block 12
☐ Approved design data and are in a condition for safe operation. ☐ Non-approved design data specified in Block 12.	Certifies that unless otherwise specified in Block 12, the work identified in Block 11 and described in Block 12 was accomplished in accordance with Title 14, Code of Federal Regulations, part 43 and in respect to that work, the items are approved for return to service.

13b. Authorized Signature:	13c. Approval/Authorization No.	14b. Authorized Signature: Gerald Young	14c. Approval/Certificate No.: CRS VY1R575K
13d. Name (Typed or Printed):	13e. Date (dd/mmm/yyyy):	14d. Name (Typed or Printed): Gerald Young	14e. Date (dd/mmm/yyyy): 09/May/2017

User/Installer Responsibilities

It is important to understand that the existance of this document alone does not automatically constitute authority to install the aircraft engine/propeller/article.

Where the user/installer performs work in accordance with the national regulations of an airworthiness authority different than the airworthiness authority of the country specified in Block 1, it is essential that the user/installer ensures that his/her airworthiness authority accepts aircraft engine(s)/propeller(s)/article(s) from the airworthiness authority of the country specified in Block 1.

Statements in Blocks 13a and 14a do not constitute installation certification. In all cases, aircraft maintenance records must contain an installation certification issued in accordance with the national regulations by the user/installer before the aircraft may be flown.

CN 62/001 Yellow Poster

6 Contrôle d'exécution des modifications et des Services Bulletins
Modification and Services Bulletin checching

Numéro *N°*	Nature de la Modification *Type of modification*	Unité ou Société d'Exécution *Performing Unit or contractor*	Date d'exécution et tampon de contrôle *Performing date Inspection stamp*
1	SB 62-011	AHI	05/09/2017 AHI 40

5 Unité ou société *Contractor*	Date *Date*	Version et N° Appareil *Aircraft Version and SN*	Fonctionnement *Operation* Support *Support*	Partiel *Partial*	Total *Total*	Attestation / *Certif.* FORTC QC
E.C	13/04/2007				0 heure	Fab/*Manuf.* Récept/*Receipt*
						Destockage/*Depreservation*
		EC 120B				V.P/*Periodic Overhaul* FORTC
EC	06/07	1491	OH		OH	Montage/*Fitting*

FICHE MATRICULE / *LOG CARD* *

Fiche n° 1
Log Card n°

~~**Fiche suivouse Matériel neuf** *Equipment record card New equipment*~~ *

1 Identification du matériel *Material identification*

Dénomination *Name*: PALE ROTOR PRINCIPAL / *Main Rotor Blades*
Nomenclature OTAN *NATO Nomenclature*:
Référence EC *EC Part Number*: C 621A 1006103
Référence fabricant *Manufacturer's Part number*:
Code OTAN fabricant *NATO Manuf code*: F 0210
Configuration de livraison *Delivery configuration*:
Version *Version*:
Numéro de série *Serial number*: 1982
Amendements *Amendements*:
Fabricant *Supplier*: EUROCOPTER
Type *Type*:

2 Marché ou commande *Contract or order*

Référence *Reference*:
Date *Date*:
Organisme émetteur *Issuing agency*:
N° ou lot *Kit n°*:
Fournisseur *Contractor*:
Adresse *Address*:

3 Garantie *Guarantee*

Matériel *Material*	Date de livraison *Date of delivery*	Durée garantie de stockage *Guarantee storage period*	Date de mise en service *Service date*	Durée Garantie de fonctionnement *Guarantee operation period*
Fab / Man				

4 Renseignements Particuliers *Special information*

Di : -1'

Tabs. : -1°/-1°/-1°/-1°/-2°/ 0°/ 0°

Réglée suivant standard
(Standard balancing)

Tab 6: + 4°
Tab 7: + 4°

FORTC QC

Limite de fonctionnement *Operating limit*	Limite de vie *Life limit*
N/A	20 000 hrs.

* Rayer la mention inutile — * Delete as necessary

5

Positions successives et opérations d'entretien et de remise en état mineures et majeures

Successive status and minor/major maintenance and overhaul operations

Unité ou Société / *Contractor*	Date / *Date*	Version et N° Appareil / *Aircraft Version and S/N*	Fonctionnement / *Operations* — Unité / *Unity* : heure — Support / *Support*	Partiel / *Partial*	Total / *Total*	Unité / *Unity* : — Support / *Support*	Partiel / *Partial*	Total / *Total*	Motif du mouvement (code symptômes) - Travaux effectués - Pièces changées / *Reason for Transfer (code and symptoms) - Accomplished work - Replaced parts* / Positions successives avant livraison avion - Voir recto tableau 5 / *Successive location before A/C delivery - See front Page table 5*
EC DS2	06/07 02/11	EC 120B 1491 1491	1506.0		2 H25 1508.2				TABS N°6 = 0° N°7 = -1° C/W 500 Insp IAW EC120b MM 62-11-00,6-1 ¶6-4 AP 3163131. J. CASTRO
— DS2 Yuma	— 25 FEB 2011	— EC120B 1490	2017.8		1508.2				INSTALLED BLADE S/N 1982 ON EC120 S/N 1490 REG # N373HS [illegible] 2739237 AP/IA
DS2 Yuma	04/11 2011	EC120 1490	2144.6		1634.4				NR Overspeed was displayed, RPM of 453 rpm during flight. An unscheduled inspection IAW MM 05-50-00, 6-1 of the controls for hard points was conducted by pilot using hydraulics and a static control check was conducted and no defects noted. BLADE S/N 1634.4 This aircraft is considered airworthy with respect with the work performed and is approved for return to service. [signature] 2822681 AP/IA
DS2 Yuma	7/16/12	EC120 1490	3001.6	857.6	2492.0				C/W 500 HR Inspection Remove Blade from Aircraft [illegible] A&P 3512893IA
DS2	8/14/12	EC120/1566	1998.0	—	2492.0				Installed Serviceable [illegible] A&P 320655
DS2/PAE	5/29/15	EC120/1566	3295.6	1297.6	3789.6				Removed from ACFT 378HS [signature] A&P 2735682 IA
AHI	14.Oct.2015	CRS#VY1R575K		TSN:	3789.6				Inspected MRB per 62-11-00. 6-1 rev. Orion 00001. Replaced lower surface PU per 62-11-00. 8-3 rev. Orion 00001. Replaced weight support cover per 62-11-00. 8-8 rev. Orion 00001, complied with SB 62-011. Refinished per 62.10.00.882 rev. MRV 003. Static balanced per 62.10.00.805 rev MRV 003. (AHI 42) Details on AHI SO#1100027 1547
PAE [illegible] Yuma	11/03/15	EC120B/1412	4202.0	0	3789.6				Installed on S/N 1412 [illegible] A&P 3603073 IA
PAE Yuma	11/12/2015	EC120B/1412	4202.0	0.0	3789.6				Removed Serviceable from ACFT W/O: 192CB-2016-R-0015 [signature] A&P 3394352 IA
PAE Yuma	3/22/2016	EC120B S/N: 1439	4000.8	0.0	3789.6				Installed under W/O 192CB-2016-R-0010 [signature] A&P 3394352 IA
PAE Yuma	4/11/16	EC120B S/N 1439	4003.2	2.4	3792.0	—	—	—	Track + Balance Complete per [illegible] S/N 1439 See W/O [illegible] Final Tab Setting Tab 6: 0.0 + Tab 7: 0.0 [signature] A&P [illegible]
PAE Yuma	5/4/2016	EC120B S/N 1439	4032.7	31.9	3821.5	—	—	—	Track + Balance Completed under W/O: 192CB-2016-R-0046-0003 Final Tab Settings: 6 = -0.5°, 7 = -0.1° [signature] A&P 3394352 IA
PAE Yuma	10/14/2016	EC120B 1439	4297.3	296.5	4086.1	—	—	—	Removed for Perforations in the Leading edge of the Stainless Steel strip near the blade Tip under W/O 192CB-2017-R-0006 [signature] A&P 3394352 IA
AHI	05/09/2017	CRS# VY1R575K	--------	-------	4086.1	-------	-------	------	Inspected blade I.A.W. 62-11-00,6-1 (Rev 000200). Replaced weight cover support per 62-11-00,8-8 SB 062-011 (Rev 00200). Paint Refinished per 62-10-00-882 MRV (Rev 003). Static balanced per MRV 62-10-00-805 (003). Replaced #1 L/E per MRV 62-10-00-831 (Rev 003), old L/E weight 234 grm, Reference SO# 11000345810 (AHI 40)
PAE / Yuma	07/06/2017	EC120B/1412	4987.7	0	4086.1	—	—	—	Installed See WO 192CB-2017-I-0041-0006 [signature] A&P [illegible]
PAE Yuma	9/23/2017	EC120B 1412	4989.3	1.6	4087.7	/	/	/	Main rotor track and balance was completed under work order: 192CB-2017-I-0046-0022; final trim tab settings for main rotor blade s/n: 1982 were as follows: tab #6: +7.6°, tab #7: +7.6° [signature] A&P 3394352 IA
/	/	/	/	/	/	/	/	/	/
Pae Yuma	12-19-17	EC120 1412	5193.6	204.3	4292.0	—	—	—	M/R T&B Performed. Final readings on TABS 6 & 7 are #6 = +0.3 degrees #7 = +0.2 degrees. [signature] A&P 2782081 IA

See Jcn 179CB-2018-I-0019.